US Department
of Transportation
Federal Aviation
Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

OMB No. 2120-0020 Exp: 01/31/2023	Electronic Tracking Number				
For FAA Use Only					

INSTRUCTIONS: Print or type all entries. See Title 14 CFR §43.9, Part 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. §44701). Failure to report can result in a civil penalty for each such violation. (49 U.S.C. §46301(a)) Nationality and Registration Mark Serial No. N2130L 1. Aircraft Model Series Make Beechcraft Name (As shown on registration certificate) Address (As shown on registration certificate) Address 10018 S RACETRACK RD CHAB DEVELOPMENT LLC 2. Owner Mineral Point State MO 63660 Country U.S.A. 3. For FAA Use Only 5. Unit Identification 4. Type Repair Alteration Unit Make Model Serial No. Beechcraft TK-25 (As described in Item 1 above) AIRFRAME X **POWERPLANT PROPELLER** Type **APPLIANCE** Manufacturer 6. Conformity Statement A. Agency's Name and Address B. Kind of Agency Adam Piepke U. S. Certificated Mechanic Manufacturer 7444 Williams In. Foreign Certificated Mechanic C. Certificate No. State MO City Bonne Terre **Certificated Repair Station** Zip 63628 Certificated Maintenance Organization Country I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge. Signature/Date of Authorized Individual Extended range fuel per 14 CFR Part 43 App. B 7. Approval for Return to Service Pursuant to the authority given persons specified below, the unit identified in item 5 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is x Approved Rejected FAA Flt. Standards Persons Approved by Canadian Maintenance Organization Manufacturer **Department of Transport** Inspector BY Other (Specify) Inspection Authorization **FAA Designee** Repair Station Designation No. 3965513 Signature/Date of Authorized Individual

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

				NO420		
				N2130L Nationality and Reg	7/8/21	Data
Removed Ione IAW	left and right	landing lights 86AK.	and replaced	with P/N 01-103	60-4596 AERO L	Date EDs lights, work

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MAJOR REPAIR AND ALTERATION

OMB No. 2120-0020 Exp: 01/31/2023	Electronic Tracking Number
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of Transportation Federal Aviati Administration	on on	(Airfran	ne, Powerplant, I	Prop	elle	er, or Appl	iance)		··	
instructions	and disposion. (49 U.S	ition of this .C. §46301	form. This report is (a))	14 CF requir	R §	§43.9, Part 43 by law (49 U.S	S.C. §44701). Fa	ailure to rep	-1 (or su ort can n	bsequent revision thereof) for each
Nationality and Registration Mark U.S.A. N2130L					Serial No.	(-25				
1. Aircraft	BEECHCRAFT					Model	STC		Series	
	1		registration certificate	•			Address (As s	-	•	certificate)
2. Owner	CHAB	DEVEL	OPMENT LLC	;	Address 10018 S RACETRACK RD City MINERAL POINT			State MO		
						· •==				ntry U.S.A.
					3. F	For FAA Use			==	
4. Ty	pe				5. l	Unit Identifica	rtion			
Repair	Alteration	Uni	t	Ma	ke			Model		Serial No.
	X	AIRFRAM	E BEECHCR	AFT			(As describe	d in Item 1 a	above)	TK-25
		POWERF	LANT	Г						
		PROPELI					·			
		APPLIAN		Type Manufacturer						
6. Conformity Statement										
A. Agency's		ddress			B. Kind of Agency					
Address 7444 WILL	PIEPKE AMS LN.				U. S. Certificated Mechanic Manufacturer X Foreign Certificated Mechanic C. Certificate No.					
	E TERRE		State MO		H	X Foreign Certificated Mechanic C. Certificated Repair Station				
Zip 63628	Co	untry U.S.A.				Certificated Maintenance Organization 396			3965513	
have b	D. I certify that the repair and/or alteration made to the unit(s) identified in item 5 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge.									
Extended range fuel per 14 CFR Part 43 App. B Signature/Date of Authorized Individual 5/25/21										
Pursuant 1	o the autho	ority given		elow,				vas inspecte		e manner prescribed by the
F/	AA Fit. Stand		Manufacturer			Maintenance Organization Persons Approved by Canadian Department of Transport				
BY	AA Designee		Repair Station	X	In	spection Author	orization	Other (Spec		
Certificate or Designation i	_{lo.} 39655	13	Signature/Date of Au	nthoriz		dividual	5/25/21			

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished (If more space is required, attach additional sheets. Identify with aircraft natio	nality and registration mark and da	te work completed.)
	U.S.A. N2130L	5/25/2021
	Nationality and Registration Mark	Date
Removed the Following instrumentation: L&R oil temp 2-804032-15 S/N ′ s25240B, 24738B; L& R TIT gauge 0146. L&R load meters P/Ns A-1155-3, S/Ns 1374, 12 boards. P/Ns 58-380051-13 S/Ns 622, 638. Tachomet Gauge P/N 5709-3006 S/N 300. Manifold Pressure Gauge P/N 5709-3006 S/N 300. Manifold Pressure Gauge P/N 5709-3006 S/N 252147 for right engine. P/N EDC-33P-4/6 S/N 252149 RFLM-4-24 S/N 25215 left fuel tanks, located behind left engine firewall. P/N 252152, RFLM-4-24 S/N 252154 for right engine and firewall. Weight and balance, and equipment list updatan electrical load analysis and determined that the totarated system capacity. Inserted FAA-APPROVED POI for CGR-30P ′ s. Inserted FAA-APPROVED POH/AFR CGR-30C. Reference CGR-30P & CGR-30C Instructions (No. 06211301 Revision: E, March 28, 2014 or latest relinstallation Instructions, Document II. Rev: B, 5/11/15. Document II 02151301 Rev. F: 5/8/15. And STC ′ s S/	/pressure/cylinder temples P/Ns EGT-100c-o-t-a 273. L&R Fuel level gauser P/N106-389005-3 S/ 2 auge P/N 6020-E67. S/N 2 following parts. Two P/N 2 N CGR-30C S/N 25214 3 PT-60ABS S/N 25215 5 EDC-33P-4/6 S/N 25215 6 ight fuel tanks, located ted to reflect this changer al continuous load does 6 H/AFM Supplement, No. AFM 6 Supplement, No. AFM 6 Supplement, No. AFM 6 Supplement, No. AFM 7 Supplement, No. AFM 8 Supplement, No. AFM 8 Supplement, No. AFM 9 Supplement, No. AFM	cluster gauges P/Ns a—i-e, S/N 's 0145, ges W/ Printed Circuit N 2504. Fuel Flow N 84353. Installed N CGR-30P, S/N 48 As Cluster gauge. 1, for left engine and 50, PT-60ABS S/N behind right engine e this date. Performed not exceed 80% of D. AFM022513, Rev. F M1030131, Rev. B For orthiness Document V CGR-30C CGR-30C tion Instructions
		;
Additional Sheets	Are Attached	

U.S. Department of Transportation Federal Aviation Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

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orm Approved MB No. 2120-0020	Electronic Tracking Number

For FAA Use Only

	Administration										
instructio	TIONS: Prints and dispos	sition of th	is form.	es. See Title 14 of This report is re	CFR §	43.9, Part 43 Ap I by law (49 U.S.	pendix B, and A C. §44701). Fai	C 43.9-1 (or silure to report	subseque can resu	ent revision thereof) for ult in a civil penalty for each	
	Nationality			nrk			Serial No.				
1. Aircraft USA N2130L					TK-25						
i. Ailcia	Make BEECH						Model 58TC		Serie BAR	=	
			-	on certificate)			Address (As shown on registration certificate)				
2. Owner		HOWAR	D			50 NORTHCREST WAY POWDER SPRINGS, GA					
							30127 USA	INGS, GA			
3. For FAA Use Only											
4.	Гуре					5. Unit Identi	ication			1	
Repair	Alteration	Ur	nit		Make			Model		Serial No.	
	X	AIRFRA	ME		1.10			in Item 1 at	ove)		
		POWER	PLANT								
		PROPE	LLER								
		APPLIANCE Manufacturer									
	1	- Indicators									
		l				6. Conformity	/ Statement				
A. Agency's Name and Address B. Kind of Agency											
Precis	ion Avion	ics Sp	ecial:	ists		U. S. Certi	ficated Mechani	С	Man	ufacturer	
	ry Whatl	ey Way				Foreign Certificated Mechanic C. Certificate No.					
Griffi 30224	•					X Certificated Repair Station PAOR-392K RADIO RADIO LIMITED INSTRUMENT					
										INSTRUMENT	
have	been made ir	n accorda	nce with	tion made to the the requiremen to the best of my	ts of P	art 43 of the U.	S. Federal Aviati	on Regulation	ns and th	se or attachments hereto at the information	
Extended			Signa	ture/Date of Author						SCOTT COLLINS	
per 14 CFI App. B	R Part 43			Scott	Coll	lin				26-August-2016	
					7. A pp	roval for Retur	n to Service				
		, ,		ns specified be Administration	,		ed in item 5 wa proved	as inspected Rejecte		manner prescribed by the	
	FAA Flt. Standa Inspector	ards	Man	ufacturer		Maintenance Orga	anization		s Approved nent of Tra	d by Canadian ansport	
BY	FAA Designee	3	Repa	air Station		Inspection Author	zation	Other (Specify	/)		
Certificate			Signa	ture/Date of Author				•		Scott Collins	
Designation No.							26-August-2016				
										-	

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

Description of Work Accomplished (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)
USA N2130L Aug-26-2016
Nationality and Registration Mark Date NSTALLED THE GARMIN GDL-88 ADSB TRANSCIEVER P/N 010-00859-00 IAW STC# SA02119SE. EQUIPMENT INSTALLED; GARMIN GDLD-88 TRANSCIEVER
1. THE FOLLOWING IS THE INSTALLATION OF A GDL-88 ADSB TRANSCIEVER SYSTEM AND IS INSTALLED IAW STC# SA02119SE. SYSTEM ACCURACY IS WITHIN SPECIFIED LIMITS PER THE GARMIN INSTALLATION MANUAL P/N 190-01122-00 REV H DATED DEC. 2014.
2. THE FOLLOWING IS THE INSTALLATION OF A GARMIN GDL-88 ADSB TRANSCIEVERS. THIS SYSTEM IS COMPATIBLE WILL ALL OTHER AIRCRAFT SYSTEMS. THE GDL 88 IS INTERFACED TO THE GARMIN GNS-430W. THIS INSTALLATION HAS BEEN EVALUATED IAW THE CRITERIA CONTAINED IN AC-20-138B. THIS INSTALLATION IS PER THE GARMIN INSTALLATION MANUAL P/N 190-01122-00 REV H DATED DEC. 2014.
3.CONTROL AND OPERATION PROCEDURES FOR THIS SYSTEM CAN BE OBTAINED IN THE GARMIN PILOTS GUIDE P/N 190-01122-0 REV H DATED DEC. 2014.
4. SERVICING OF THE GDL-88 SYSTEM IS NOT APPLICABLE.
5. PERIODIC MAINTENANCE INSTRUCTIONS CAN BE FOUND IN THE GDL-88 INSTALLATION MANUAL P/N 190-01122-00 REV H DATEI DEC. 2014.
6. TROUBLESHOOTING INFORMATION CAN BE FOUND IN THE GDL-88 INSTALLATION MANUAL P/N/ 190-01122-00 REVH DATED DEC 2014.
7. FOR REMOVAL AND REPLACEMENT INFORMATION REFER TO THE GDL-88 INSTALLATION MAMUAL P/N 190-01122-00 REV H DATED DEC. 2014.
8. DIAGRAMS FOR ACCESS PLATES ARE NOT REQUIRED/NONE APPLICABLE.
9. THERE ARE NO SPECIAL INSPECTIONS FOR THIS SYSTEM.
10. APPICATION OF PROTECTIVE TREATMENTS IS NOT APPLICABLE.
11. DATA FOR STRUCTUAL FASTENERS CAN BE FOUND IN THE GDL-88 INSTALLATION MAMUAL P/N 190-01122-00 REV H DATED DEC. 2014.
12.THERE IS NO SPECIAL TOOLS REQUIRED.
13. COMMUTER AIRCRAFT INFORMATION IS NOT APPLICABLE.
14. THERE ARE NO RECOMMENDED OVERHAUL PERIODS FOR THIS SYSTEM.
15. THERE ARE NO ADDITIONAL AIRWORTHINESS LIMITATIONS.
16. REVISION: TO THE INSTRUCTIONS FOR CONTINUED AIRWORTHINESS ARE TO BE SUBMITTED VIA LETTER FORM AND REVISE F.A.A. FORM 337 TO LOCAL FSDO FOR APPROVAL.
END
☐ ADDITIONAL SHEETS ARE ATTACHED

U.S. Department of Transportation Federal Aviation Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No.2120-0020 For FAA Use Only

Office Identification

e								
Alteration								
x								
6. Conformity Statement								
A&P143423552								
ents								
J. Price								
d by								
n								

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

Removed existing rudder mount Grimes oscillating beacon and installed a Whelen recommended solid-state halogen flasher, Model 9004457 red, FAA/TSO-C96A P/N 01-0790044-57. Removed existing Grimes oscillating beacon on underside of the fuselage and installed a Whelen recommended solid-state halogen flasher, Model 9004458 white, FAA/TSO-C96A P/N 01-0790044-58. Installation per Whelen Installation Guide for model 90044-(), using existing mounting holes and adapter plates.

The electrial load is less than the original installed equipment.

All avionics systems were checked for EMF interference.

No weight and balance change, equipment list revised.

ICA CHECKLIST

- 1. Introduction: Aircraft registration No N2130L, Manufacture: Raytheon/Beech, Model 58TC, S/N TK-25.
- 2. Discription: Whelen Flashing Anti-Collision Light Assembly, TSO-C96A.
- 3. Control/Operation Information: Uses existing mounting holes, wires, switches and breaker.
- 4. Service Information: Use Beech and Whelen Maintenance Manuals.
- 5. Maintenance Instructions: Use Beech and Whelen Maintenance Manuals. Inspect at normal Inspection periods, 100 Hr/Annual.
- 6. Trouble Shooting Information: Use Beech and Whelen Maintenance Manuals.
- 7. Diagrams: Whelen Installation Guide Model 90044-().
- 8. Removal and Replacement Information: Use Beech and Whelen Maintenance Manuals.
- 9. Special Inspections Requirements: N/A
- 10. Application of Protective Treatments: When necessary, waterproof the flasher base with

RTV sealant. 11. Data Units Installed with standard AN/M 12. Special Tools: N/A 13. Commuter Caterogy: N/A 14. Recommended Overhaul Period: No ad 15. Airworthiness Limition Section: N/A	ditional overhaul time limitations.
 Revision: A letter will be submitted to th 337 and revised ICA. 	e local FSDO with a copy of the revised FAA form
*********	******* NOTHING FOLLOWS ************************************
	• •
	Additional Sheets Are Attatched

US Department of Transportation

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form /	Appr	oved	ı
OMB	No.	2120	0-0020

For FAA Use Only

Office Identification

INSTRUCTIONS: Print or type all entries. See FAR 43.9 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act of 1958).

		such violation					of 1958).	, and				portainy 1100		
		Make Beech					Model							
1. A	Aircraft	Serial No. TK-25					58TC Nationality and Registration M N2130L							
2. C	Owner	Name (As shown on registration certificate A36 Leasing Corp.						35	stration Suite 10-49					
					3	. For	FAA Use Onl	у						
				4. Uni	it Identific	catio	n					5. Type		
	Unit		Make	9		l	Model		Seri	al No.		Repair	Alteration	
AIR	RFRAME	(As descri					in Item 1 above	e) ~~~	~~~~				х	
PO	WERPLANT									,				
PR	OPELLER													
API	PLIANCE	Type Manufacturer												
^ ,	Vacacida NI				6.		ormity Statem				<u> </u>	velificate \$1-	<u> </u>	
A. F		e and Address SON AVIATION		INC		B. Kind of Agency C. 9 U.S. Certified Mechanic						. Certificate No.		
		CER COUNT	,				Foreign Certif	ified M		CRS EHHI				
		TON, NJ 08				X	Certificated R Manufacturer			Airframe C Radio Clas				
	have been ma	he repair and/o	nce w	vith the require	ments of	Part 4	entified in item 4 43 of the U.S. I	4 abov						
Dat	e .	ein is true and o	corre	ct to the best (or my kno		ge. nature of Autho	orized	Individual					
4-3-07					Alex	xander Eder	بی	lezam	la	مر	2			
Pi	rsuant to the	authority given	pere	ons specified h			for Return to identified in ite			in the ma	nner n	rescribed by	the	
Pursuant to the authority given persons specified below, the Administrator of the Federal Aviation Administration and is							APPRO			IECTED				
BY	I E	AA Fit. Standards Manufacturer				Inspection Au	uthoriz	ation	Other (S	specify,	')			
		A Designee X Repair Station					Person Appro Canada Airwo	orthine	ess Group				***************************************	
Date of Approval or Rejection Certificate or Designation No. CRS EHHR538D					Signature of Authorized Individual Alexander Eder Alexander FAA Form 337 (12-88)									
Alexander Edel Williams								(12-08)						

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets, Identify with aircraft nationality and registration mark and date work accomplished.) Validated that the previous installation of one GNS-530 was installed IAW Garmin instructions and approved via an FAA stamped field approval document on FAA Form 337. Verified this aircraft and all interfaced equipment are covered under the STC AML. The unit was removed and upgraded to a GNS-530W unit. The existing location of the unit was determined to meet the field of view requirements without the need for external annunciation. Verified that RG-142 antenna cable was used for the GPS antenna. The existing GA-56 antenna interfaced with the GNS-530 was removed and replaced with a GA-35 GPS antenna using the approved mounting provisions of the previous installation. Sealed the GPS antenna with RTV.

- A. Removed a Garmin GA-56 antenna P/N 011-00134-00 and installed a new GA-35 GPS/WAAS antenna P/N 013-00235-00 using provisions left behind from the standard antenna IAW Garmin upgrade manual P/N 190-00357-06 Rev A and STC SA01933LA.
- B. Removed Garmin GNS-530 P/N 011-00550-10 and installed Garmin GNS-530W P/N 011-01064-40 S/N 78485419, using provision left behind from the standard 530 unit. Installation done IAW Garmin upgrade installation manual P/N 190-00357-06 Rev A and STC SA01933LA.
- C. The GNS-530W was configured identical to the original 530 unit. Each interface was checked out IAW the 530W Installation Manual P/N 190-00357-02.
- D. Removed the Aircraft Flight Manual Supplement for the GNS-530 and installed a GNS-530W AFMS P/N 190-00357-63, FAA Approved dated 12/21/2006 into the Aircraft Flight Manual.
- E. Updated aircraft equipment list, weight and balance change is negligible.
- F. Reference Ronson Aviation Inc. W/O 323760.

nstructions for Continued Airworthiness (ICA):
Reference Attached document P/N 190-00356-65
END

U.S. Department of Transportation

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved
OMB No.2120-0020
For FAA Use Only

Office Identification

Administration											
instructions and	 Print or type all e disposition of this for or each such violation 	rm. This report	is required	d by	law (49 U.S.	C. 142	AC 43.9-1 (21). Failur	or subseque to report of	ent rev can res	rision thereof) ult in a civil p	for enalty not to
1. Aircraft	Make BEECH			Model 58TC							
	Serial No. TK-25					Natio	-	egistration Ma	rk		
2. Owner	Name (As shown or A36 LEASING Co			351	1 SILVERS	vn on registrati IDE RD STE , DE 19810-	E 105	īcate)			
	<u></u>		3. F	or l	FAA Use Only	,					
			- 1								
		4. Uni	t Identific							5. T	уре
Unit	Mak	е			Model		\$	Serial No.		Repair	Alteration
AIRFRAME		(A	ed in	n item 1 above	-)		·			X	
POWERPLANT		·	. • .								
PROPELLER											
APPLIANCE	Туре										
74 CDATOL	Manufacturer		TI.								
			6. Coi	nfo	rmity Stateme	ent	·			1, , , , , , , , , , , , , , , , , , ,	I
A. Agency's Na	me and Address				B. Kind of A				C. C	ertificate No.	
William J. Price Hortman Aviation				X		Certificated Mechanic				143423552	
9800 Ashton Rd	,			\vdash	Foreign Certificated Mechanic						
Philadelphia, PA	. 19114			-	Certificated Repair Station Manufacturer						
hereto have	the repair and/or al been made in acco furnished herein is t	rdance with the	requireme	nts	entified in item of Part 43 of t	he U.	ove and de S. Federal	escribed on Aviation Re	the revegulation	verse or attac	hments he
Date March 3, 2006			Sig	nati	ure of Mitboriz	ed In	dividual	nd		Wil	liam J. Price
					or Return To	<u> </u>					
the Admin	to the authority give istrator of the Fede	n persons speci al Aviation Adm	fied below, inistration	the	e unit identified d is 🕢 APPRO	d in ite	em 4 was i	REJECTE	D	anner prescr	ibed by
BY Inspe	Fit. Standards ector	Manufacturer		X	Inspection /			Other (S)	pecify)		
	N Designee	Repair Station		_	Person Appove	thiness	Group)			
Date of Approva	or Rejection	Certificate or Des	signation No.		Signature	OT AU	horized in	gividua	(, sam	iom I Brios
March 3, 2006 FAA Form 337	(12-88)	143423552			Wil	n	any	- 1	7	· VVIII	iam J. Price

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished (If more space is required, attach additional sheets. Id	dentify with aircraft r	nationality and registra	tion mark and date work completed.)								
Installation of Teledyne Continental Motors TSIO-540-WB engines and increase maximum takeoff and landing weights to 6200 pounds in accordance with CenTex Aerospace, Inc. Drawing Number CTA58-1001, dated January 12, 2000, or later FAA approved revision. FAA approved Airplane Flight Manual Supplement Number AFMS 1000, dated September 7, 2000, or later FAA approved revision was installedin in Flight Manual. STC SA09618SC.											
**********	NOTHING FOLLO)WS ************	*****								
	2.7%.A		e geografie								
	V = 1 = 1 × 10 × 10 × 10 × 10 × 10 × 10 ×	, d									
e de la companya della companya dell	u kanala Wasan		en ja tulkaan oli 1990. Heriotoidaksi kan pinnin en 1990. Herioto Tooloo kan tootoo kan 1998.								
		· .	magaya sa waseman ka wasa ka ƙasar								
	• '										
* .											
	w.										
			,								
,	Additional Sheets Ar	re Attatched									

Muited States Of America

Bepartment of Transportation - Federal Abiation Administration

Supplemental Type Certificate

Number SA09618SC

This Certificate issued to

CenTex Aerospace, Inc. 7805 Airport Drive P. O. Box 5803 Waco, TX 76708

cortifies that the change in the type design for the following product with the limitations and conditions therefor as specified hereon meets the aircorthiness requirements of Part 23 of the Federal Aviation Regulations.

Original Product Type Certificate Number:

A23CE

Make:

Beech

Model:

58P; 58TC

STC SA09618SC AND SUPPLEMENT AFMS 10000 ARE PROVIDED FOR MODIFICATION OF ONLY THE AIRCRAFT LISTED BELOW UNAUTHORIZED USE FOR OTHER AIRCRAFT IS THEFT AND WILL BE SUBJECT TO PROSECUTION BY LAW:

AIRCRAFT SERIAL NO. TK-25

REGISTRATION NO. N2130L

Description of Type Design Change: Installation of Teledyne Continental Motors TSIO-520-WB engine and increase maximum takeoff and landing weights to 6200 pounds in accordance with CenTex Aerospace. Inc. Drawing Number CTA58-1001, dated January 12, 2000, or later FAA approved revision. FAA approved Airplane Flight Manual Supplement Number AFMS 1000, dated September 7, 2000, or later FAA approved revision is required.

Limitations and Conditions. Compatibility of this design change with previously approved modifications must be determined by the installer. If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

This certificate and the supporting data which is the basis for approval shall remain in offect until surrendered, suspended, revoked or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of upplication January 12, 2000

Dute of issuance: October 20, 2000

Date reissued:

Date amended:

By direction of the Administrator

(Signature

S. Frances Cox

Manager, Special Certification Office

Southwest Region

(Title)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years. or roth.



AFMS 1000

FAA APPROVED AIRPLANE FLIGHT MANUAL SUPPLEMENT FOR STC SA09618SC TSIO-520-WB ENGINE INSTALLATION IN BEECH MODELS 58P and 58TC AIRPLANES

Registration No. N2130L
Serial. No. TK-25
Model No. 58TC

This supplement is FAA approved and must be attached to the FAA-approved Airplane Flight Manual when the aircraft has been modified by the installation of a Teledyne-Continental Motors-TSIO-520-WB engine in accordance with S.T.C. SA09618SC. The information contained in this document supplements or supersedes the Airplane Flight Manual only in those areas listed herein. For limitations, procedures, and performance not contained in this supplement, consult the Airplane Flight Manual.

FAA APPROVED

S. Frances Cox, Manager

Special Certification Office, ASW-190

Federal Aviation Administration Fort Worth, Texas 76193-0190

DATED September 7, 2000

STC SA09618SC AND SUPPLEMENT AFMS 1000
ARE PROVIDED FOR MODIFICATION OF ONLY
THE AIRCRAFT LISTED BELOW. UNAUTHORIZED
USE FOR OTHER AIRCRAFT IS THEFT AND WILL
BE SUBJECT TO PROSECUTION BY LAW.
AIRCRAFT SERIAL NO: TK-25
REGISTRATION NO: N2130L

CENTEX AEROSPACE INC. P. O. Box 5803 Waco, Texas 76708

REVISION	CHANGE DESCRIPTION	BY	APPROVAL	DATE
IR	initial release	GB	G.R.B.	01/12/ 2000
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PROPRIETARY INFORMATION

THIS DOCUMENT CONTAINS TRADE SECRET INFORMATION OWNED BY CENTEX AEROSPACE

INC. UNAUTHORIZED DISCLOSURE OR USE OF THE INFORMATION CONTAINED IN THE

DOCUMENT TO GROUPS OR INDIVIDUALS WILL BE TREATED AS THEFT OF A TRADE SECRET.

DISCLOSURE OR USE OF THE INFORMATION CONTAINED IN THIS DOCUMENT TO SPECIFIC

GROUPS OR INDIVIDUALS IS AUTHORIZED ONLY

BY MEANS OF A LETTER OF AUTHORIZATION

SIGNED BY THE PRESIDENT OF CENTEX

AEROSPACE INC.

DWG ELIGIBILITY:

BEECH MODEL 58P SERIAL NOS: TJ-3 THROUGH TJ-168
BEECH MODEL 58TC SERIAL NOS: TK-1 THROUGH TK-84

NOTES

- A. PERFORM ALL PROCEDURES PER THE CURRENT BEECH SHOP MANUAL, MODEL 58 AND/OR TELEDYNE CONTINENTAL MOTORS MAINTENANCE MANUAL (MODEL TSIO-520); UNLESS OTHERWISE NOTED.
- B. UNLESS OTHERWISE LISTED HEREIN, USE HARDWARE AS SPECIFIED IN THE CURRENT BEECH PARTS CATALOG.
- C. ALL PARTS OTHER THAN NEW MUST BE SERVICEABLE, CLEANED, DIE-CHECKED, AND INSPECTED PRIOR TO INSTALLATION.
- D. INSTRUMENT RE-MARKING MUST BE PERFORMED BY AN FAA AUTHORIZED REPAIR STATION.

PREPARATION PROCEDURE:

- 1. REMOVE MANIFOLD PRESSURE GAGE, AIRSPEED INDICATOR, AND TACHOMETER (IF REQ'D).
- 2. INSTALL TAIL STAND (OR EQUIVALENT) ON AIRCRAFT PRIOR TO ENGINE REMOVAL.
- 3. INSPECT ENGINES AND PROPELLERS FOR ELIGIBLE MODEL (see page no. 2 this dwg). IF ENGINES ARE TCM MODEL TSIO-520-LB AND WILL BE MODIFIED TO TSIO-520-WB, PERFORM MODIFICATION PER TCM SERVICE BULLETIN NO. 75-6. IF ENGINES ARE TCM MODEL TSIO-520-L, REPLACE WITH ELIGIBLE MODEL ENGINES.

MODIFICATION PROCEDURE:

- 2. CHECK AIRSPEED INDICATOR MARKINGS. REMARK AS SHOWN:

MARKING	IAS VALUE (knots)	SIGNIFICANCE
White Arc	78-143	Full Flap Operating Range
White Triangle	177	Maximum Flap Approach Position 15 °
Blue Radial	115	Single-Engine Best Rate-of-Climb
Red Radial	81	Minimum Single-Engine Control (V _{MCA})
Green Arc	84-196	Normal Operating Range
Yellow Arc	196-235	Operate With Caution, Only In Smooth Air
Red Radial	235	Maximum Speed For All Operations (Never Exceed)

CenTex Aerospace Inc

P.O. BOX 5803 - WACO, TX 76708 - 254/752-4290

TITLE: BEECH 58TC &	58P ENGINE UPGRAD	DE DWG	NO: CTA58-1001	REV: -
BY: G BARNES	SCALE: NONE	DATE: 01/12/00	CKD BY:	PAGE 1 of 2

4. INSTALL TSIO-520-WB ENGINES OR VERIFY MODIFIED PER TCM SB75-6.

5. INSTALL PROPELLERS OR VERIFY MODEL (see eligible propeller models listed below; matched pairs only):

Eligible propeller models

Propeller Limits

McCauley 3AF32C511-/X//G-82NEA-4 P/N P5115358-01 or P/N P5115358-0152 Diameter: 78 in. (77.5 in. min. allowed for repair)
Pitch setting (at 30 in. Sta.): low 16.1° ±.2°

high 82.5° ±.5°

Hartzell
PHC-J3YF-2F/FC7663DR or
PHC-J3YF-2F/FC7663DR or
PHC-J3YF-2UF/FC7663DR or
PHC-J3YF-2UF/FC7663DRB or
PHC-J3YF-2UF/FC7663DRK

Diameter: 78 in. (77.5 in. min. allowed for repair) Pitch setting (at 30 in. Sta.): low 15.3°

high 84°

6. INSTALL PROPELLER GOVERNOR; Eligible part numbers: Beech 96-380030 or Beech 96-106-389001 or equivalent Woodward governor.

PREFLIGHT PREPARATION PROCEDURE:

- 1. SETUP ENGINE FUEL INJECTION SYSTEM AND PROPELLER GOVERNOR.
- 2. CHECK POWERPLANT CONTROL SYSTEMS, ELECTRICAL SYSTEMS, AND ENGINE INSTRUMENTS FOR PROPER OPERATION.
- 3. UPDATE AIRPLANE WEIGHT & BALANCE RECORDS. COMPUTE NEW EMPTY WEIGHT, MOMENT ARM, AND USEFUL LOAD BASED ON NEW MAX. TAKEOFF WEIGHT OF 6200 POUNDS.
- 4. COMPLETE AIRCRAFT LOGBOOK ENTRIES AND FORM 337.
- 5. ATTACH CENTEX AEROSPACE INC. AIRPLANE FLIGHT MANUAL SUPPLEMENT P/N AFMS 1000 TO THE AIRPLANE FLIGHT MANUAL.

INSTRUCTIONS FOR CONTNUED AIRWORTHINESS

This modification does not alter the manufacturer's instructions for continued airworthiness. Follow instructions contained in the Beech Shop Manual for the model 58P or model 58TC, as applicable.

PROPRIETARY INFORMATION

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CenTex Aerospace Inc

P.O. BOX 5803 - WACO. TX 76708 - 254/752-4290

						
TITLE: BEECH 58TC &	58P ENGINE	DWG	NO: CTA58-1001	REV: -		
	SCALE:	NONE	DATE:	01/12/00	CKD BY:	PAGE 2 of 2

US Department of Transportation

Federal Aviation
Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No. 2120-0020

For FAA Use Only

Office Identification

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Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

Dated 01-23-06 Engine Previous Total:1600.0 (Est.)

Installed locknuts and cylinder brackets on cylinder attachment studs per Dwg. 2197, Rev. AF dated 3/11/04 I/A/W STC SE3630SW.

Installed Slick pressurized magneto system per Dwg. 1069, Rev. G dated 2/12/02 I/A/W STC SE5535SW.

Installation mechanic must complete Block 1 and 2 on reverse side and mail one copy to their local FSDO.

Negligible weight and balance change.

Customer furnished with FAA approved Overhaul and Parts Manual Supplements with Instructions for Continued Airworthiness for all alterations.

Pertinent details of the above installations are on file under Project No. 2549/24325.

END	
	April 1980 - San

United States Of America

Bepartment of Transportation - Nederal Abiation Administration

Supplemental Type Certificate

Number SE3630SW

This Certificate issued to

RAM Aircraft, Limited Partnership 7505 Karl May Drive Waco, TX 76708

Waco, IX 76708
certifies that the change in the type design for the following product with the limitations and conditions
therefor as specified hereon meets the aircoorthiness requirements of Purt 13 of the Civil Air Regulations.
Original Product Type Certificate Number ESCE
Mbake: Teledyne Continental Motors.
Model: TSIO-520 -B,C,D,E,G,H,J,K,L,M,N,P,R,T,BB, EB,JB,KB,LB,NB,VB,WB, and AF
Description of Type Design Change: Crankcase modifications in accordance with RAM Aircraft Corporation Drawing No. 1157, Rev. E, dated July 9, 1986, or later FAA approved revision.
Similations and Conditions. The following supplements or later FAA Approved revisions are required:
1. Overhaul Manual Supplement dated June 1986. 2. Parts Manual Supplement dated June 1986.
Compatibility of this design change with previously approved modifications must be determined by the installer. If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

person written evidence of that permission.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, recoked or a termination data is otherwise established by the Bedministrator of the

Federal Sociation Solministration.

Date of application: June 05, 1985

Dalo reissued: October 8, 2001

Date of issuance: July 10, 1986

Date amended:



By direction of the Administrator

(Signature)
S. Frances Cox, Manager
Special Certification Office,
Southwest Region

(Title)

Bepartment of Transportation - Nederal Abiation Administration

Supplemental Type Certificate

Number SE5535SW

This Certificate issued to

RAM Aircraft, Limited Partnership 7505 Karl May Drive Waco, TX 76708

certifies that the change in the type design for the following product with the limitations and conditions therefor as specified hereon meets the airworthiness requirements of Part 13 of the Civil Air Regulations.

Original Product Type Certificate Number: E8CE

Make: Teledyne Continental Motors

Model: TSIO-520-L, -IB, -WB

Description of Type Design Change:

Install Slick Pressurized Magneto System in accordance with Drawing 1069, Rev. A, dated 6/28/83, or later FAA approved revision.

Limitations and Conditions.

Compatibility of this design change with previously approved modifications must be determined by the installer. If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked or a termination date is otherwise established by the Redministrator of the

Federal Aviation Administration.

Date of application: June 22, 1983

Dulo of issuance: September 15, 1983

Date amended: January 9, 1984

Judd: October 08, 2001

Revision 1

By direction of the Administrator

S. Frances Cox, Manager

S. Frances Cox, Manager Special Certification Office,

Southwest Region

(Title)



US Department of Transportation Federal Aviation

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No. 2120-0020

For FAA Use Only

Office Identification

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Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(if more space is required, attach additional sneets, identity with aircraft hattoriality and registration mark and date work completed.)
Installed Garmin GDL 49 Datalink Weather System.
Installation work was performed in accordance with the instructions contained in the Garmin GDL 49 Installation Manual# 190-00231-00 Rev A dated May 02, recommendations contained in FAA Advisory Circulars 43.13.1B, Chapters 11 and 12, and 43.13.2A, Chapters 1, 2, 3, and current accepted industry standards.
The GDL 49 is interfaced to the GNS 530.
An electrical system load analysis was conducted to insure that the load does not exceed 80% of the charging system capacity.
This installation has been functionally tested and found to operate normally and is in compliance with FAR 23.1301.
These units will be maintained in accordance with the manufacturer's maintenance instructions and inspected in accordance with FAR Part 43 Appendix D(j).
The aircraft Weight and Balance and Equipment List have been updated to reflect these changes. An entry was made in the airframe logbook referring to alterations detailed on this FAA Form 337.
A copy of the GNS 400/500 Series Weather Datalink Pilot's Guide PN 190-00231-05 Rev A or later, and an FAA Approved GDL 49 Aircraft Flight Manual Supplement dated <u>>/20/05</u> must be in the aircraft and immediately available to the flight crew , whenever flight is predicated on use of the system.
Instructions For Continued Airworthiness:
ICA Checklist; Reference FAA Order 8300.10 Change 15.
Item 1and 2: Introduction and Description of the Alteration are adequately detailed above. Item 3. Controls, Operation Information: Operation of the Systems is described in the FAA Approved Aircraft Flight Manual Supplement. Item 4. Servicing: Servicing of the installed equipment is required if equipment failures occur, or mandatory service
bulletins are issued. Servicing must be accomplished by an appropriately rated facility.
Item 5. Maintenance Instructions: To insure integrity, the system must be checked in accordance with the manufacturer's post-maintenance procedures following any maintenance performed on the system. This functional
check must follow the maintenance and precede any IFR operational use of the system.
Item 6. Troubleshooting: Appropriately rated facilities will be capable of troubleshooting the installed equipment with manufacturer documents.
Item 7. Removal and Replacement: Appropriately rated facilities will be capable of removing and Installing equipment
with manufacturer documents Item 8-15: N/A
Item 16: To revise this ICA, submit a letter, a copy of the FAA 337, and the revised ICA to the local FAA FSDO. After
acceptance of the revision, a maintenance entry will be made identifying the revision, its location, and the date on the FAA 337 form.
Details of work performed are on file under work order #1207.
E N D

Additional Sheets Are Attached

US Department of Transportation

Federal Aviation Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No. 2120-0020

For FAA Use Only

Office Identification—A - I

INSTRUCTIONS: Print or type all entries. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) for instructions and disposition of this form. This report is required by law (49 U.S.C. 1421). Failure to report can result in a civil penalty not to exceed \$1,000 for each such violation (Section 901 Federal Aviation Act of 1958).

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Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets, Identify with aircraft nationality and registration mark and date work completed.)

Cessna S/N TK-25 N2130L Dated 7-24-02 Engine Previous Total: 3033.5

Engine crankcase modified per Dwg. 1157, Rev. AE dated 2/1/00 I/A/W STC SE3630SW.

Installed Slick pressurized magneto system per Dwg. 1069, Rev. F darted 11/5/92 I/A/W STC SE5535SW.

Installed spring loaded induction hose clamps per Dwg. 1171, Rev. B dated 5/24/00 I/A/W STC SE3632SW.

Installation mechanic must complete Block 1 and 2 on reverse side and mail one copy to their local FSDO.

Negligible weight and balance change.

Customer furnished with FAA approved Overhaul and Parts Manual Supplements for all alterations.

Pertinent details of the above installations are on file under Project #303/2479.

END	

US Department of Transportation **Federal Aviation** Administration

MAJOR REPAIR AND ALTERATION (Airframe, Powerplant, Propeller, or Appliance)

Form Approved OMB No. 2120-0020

For FAA Use Only

Office Identification

JLL AEA-FSDO-23

and disposition	NS: Print or type all en of this form. This form riolation (Section 901 Fo	is required by law (4	\$9 U.S.C	3 App 2. 142	pendix B, and A 1). Failure to re	C 43.9-1 (or eport can resu	subsequent revision ult in a civil penalty	on thereof) for not to exceed	r instruction d \$1,000	าร	
	Make Beechcr					Model 58TC					
1. Aircraft	Serial No. TK25			Nationality and Registration Mark 2130L							
2. Owner	Name (As shown on A36 Lea	registration certificat sing Corp			Address (As shown on registration certificate) 3511 Silverside Rd STE 105 Wilmington DE 19810-4902						
applicable airwords approved only aircraft subject a person author	ed herein complied with thiness requirements and y for the above described to conformity inspection ized in FAR-43.7		nly for model	·	AA Use Only A-FSDO-	92			<u>,</u>		
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FAA Form	337 (12-88)										

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. Description of Work Accomplished

(If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

Removed: Collins VHF 251 Com, Collins VIR 351 Nav, Collins ADF 650, Foster RNAV 511 Installed: Garmin GNS 530 VOR/LOC/GS/GPS/COMM System, RC Allen Electric Gyro

The units were mounted in the aircraft manufacturer radio stack, and instrument panel. The GNS 530 is connected to the HSI and autopilot, and is connected to channel the DME.

This installation was accomplished in accordance with Garmin GNS 530 Installation Manual #190-00181-02 Rev D dated April 2001, AC43.13-1B, chapters, AC43.13-2A, chapters 1, 2, 3, FAR 23.1431, and current accepted industry standards.

The GNS unit was connected in parallel to the existing encoding altimeter. The altitude reporting equipment was tested in accordance with FAR 91.413, and Part 43 Appendix "F".

An electrical system load analysis was conducted to insure that the load does not exceed 80% of the charging system capacity.

This installation has been functionally tested and found to operate normally and is in compliance with FAR 23.1301.

These units will be maintained in accordance with the manufacturer's maintenance instructions and inspected in accordance with FAR Part 43 Appendix D(j).

The aircraft Weight and Balance and Equipment List have been updated to reflect these changes. An entry was made in the airframe logbook referring to alterations detailed on this FAA Form 337.

The Garmin GNS530 is TSO approved under category C129-A1 as a Global Positioning System. The system has been previously approved for VFR/IFR Enroute, Terminal Area, and Non-Precision Approach Navigation under STC SA00864WI in a Piper PA32, which is the basis for this request for approval for this aircraft. This installation was tested IAW AC20-138, Paragraph 8.c.(2).

GPSs placarded "GPS NOT APPROVED FOR IFR".

NOTE: The VFR only placard is a temporary placard and will be replaced with a "IFR Approved for Enroute, Terminal, and Non-Precision Approach" placard following successful completion of the flight evaluation program in accordance with AC 20-138 paragraph 8 and flown in accordance with FAR 91.407(b). The pilot must enter the results of the flight evaluation, the date, the pilot signature, and the pilot certificate number in the airframe logbook in accordance with FAR 91.107(b), which will then certify the GPS installation for IFR use for enroute, terminal and non-precision approach.

ADDITIONAL NOTE: This VFR/IFR 337 approval is only valid if the aircraft is flight evaluated on the first flight immediately following the GPS installation.

A copy of the Garmin GNS530 is in the aircraft and must be immediately available to the pilot whenever navigation is predicated on the use of the GPS System.

An FAA approved Aviation Flight Manual Supplement dated 6/20102 is in the aircraft and must be immediately available to the pilot whenever navigation is predicated on the use of the Garmin GPS System.

Instructions for Continued Airworthiness:

To insure integrity, the system must be checked in accordance with the manufacturer's post-maintenance procedures following any maintenance performed on the system. This functional check must follow the maintenance and precede any IFR operational use of the system.

Details of work performed are on file under work order #1181	•
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Federal Aviation Administration

MAJOR REPAIR AND ALTERATION
(Airframe, Powerplant, Propeller, or Appliance Follows)

(Airframe, Powerplant, Propeller, or Appliance Follows)

(CE-01)

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Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

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DEPARTMENT OF TRANSPORTATION Form Approved
Budget Bureau No. 04-R060.1 .IAN 2 0 1995 FEDERAL AVIATION ADMINISTRATION FOR FAA USE ONLY MAJOR REPAIR AND ALTERATION ACE-FSDO-61 (Airframe, Powerplant, Propeller, or Appliate MOINES, IONA OFFICE IDENTIFICATION and **CE-01** INSTRUCTIONS: Print or type all entries, for instructions and disposition of this form. See FAR 43.9, FAR 43 Appendix B, and AC 43.9-1 (or subsequent revision thereof) MAKE MODEL 58TC Baron BEECHCRAFT 1. AIRCRAFT SERIAL NO. NATIONALITY AND REGISTRATION MARK TK-25 N2130L NAME (As shown on registration certificate) ADDRESS (As shown on registration certificate) 2. OWNER MARK GLINZ R.R. 2, Box 81 Bottineau, North Dakota 3. FOR FAA USE ONLA 4. UNIT IDENTIFICATION 5. TYPE UNIT MAKE MODEL SERIAL NO. REPAIR AIRFRAME (As described in item 1 above) XX**POWERPLANT** PROPELLËR APPLIANCE MANUFACTURES 6. CONFORMITY STATEMENT AGENCY'S NAME AND ADDRESS B. KIND OF AGENCY C. CERTIFICATE NO. U.S. CESTIFICATED MECHANIC DAVE HEISTERKAMP AIRCRAFT SALES, INC Limited FOREIGN CERTIFICATED MECHANIC P.O. Box 13 - Municipal Airport Beechcraft CERTIFICATED REPAIR STATION Onawa, Iowa 51040 VXJR876L MANUFACTURER I certify that the repair and/or alteration made to the unit(s) identified in item 4 above and described on the reverse or attachments hereto have been made in accordance with the requirements of Part 43 of the U.S. Federal Aviation Regulations and that the information furnished herein is true and correct to the best of my knowledge. DATE SIGNATURE OF AUTHORIZED INDIVIDUAL 11-30-94 7. APPROVAL FOR RETURN TO SERVICE Pursuant to the authority given persons specified below, the unit identified in item 4 was inspected in the manner prescribed by the Administrator of the Federal Aviation Administration and is APPROVED REJECTED FAA FIT, STANDARDS OTHER (Specify) MANUFACTURER INSPECTION AUTHORIZATION BY CANADIAN DEPARTMENT OF TRANSPORT INSPECTOR OF AIRCRAFT FAA DESIGNEE REPAIR STATION

FAA Form 337 (7-67)

DATE OF APPROVAL OR REJECTION

11-30-94

☆ U.S. Government Printing Office 1977—772-648/141

SIGNATURE

OF AUTHORIZED INDIVIDUAL

CERTIFICATE OR

VXJR876L

DESIGNATION NO.

(8320)

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. DESCRIPTION OF WORK ACCOMPLISHED (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

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ADDITIONAL SHEETS ARE ATTACHED

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Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8, DESCRIPTION OF WORK ACCOMPLISHED (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

Removed Apollo 604 loran with Al6 antenna and installed GPS155 in same panel location using existing NAT RSOB transfer switch to hook to HSI and existing PB08 switch/annunciator for Nav 1/GPS. Installed LL08 annunciator for GPS MSG/ GPS WPT. Installed mfg supplied GA56 GPS ant using existing ant location. Installation done per mfg installation manual PN 190-00065-02 Rev E dated March 1994 and AC43.13-2A Chp 2 par 21, 22 6 23 and chapter 3 as applicable. Installation done in compliance with AC20-138. This is done as a follow on to STC SA 00133WI.

Flight test dated 12-19-94 found system to meet accuracy requirements of AC20-138 and is hereby certified for IFR enroute and terminal. Weight & balance amended, logbook entry made, FAA approved flight manual supplement provided and operators handbook supplied. Placarded panel GPS not approved for Instrument Approaches.

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DADDITIONAL SHEETS ARE ATTACHED

PU.S. GOVERNMENT PRINTING OFFICE: 1986-659-121/40260

Waited States of America

Department of Transportation—Acderal Aviation Administration

# Supplemental Type Certificate

Number SA00133WI

This corlificate issued to

GARMIN International 9875 Widmer Road Lenexa, KS 66215

cortifies that the change in the type design for the following product with the limitations and conditions.

Therefor as specified horson meets the sirverthiness requirements of Port 3 of the Civil Air Regulations.

Original Product - Typo Cartificato Number: 2A3

Make: Mooney

Model: M20J

Description of Type Design Change: Installation of GARMIN Global Positioning System GPS 155 in accordance with (1) GARMIN Master Drawing List, GPS 155 Installation in Mooney Model M20J, Rev F, dated February 7, 1994, and (2) FAA Approved Airplane Flight Manual Supplement (AFMS) for Mooney M20J with GARMIN GPS 155 Global Positioning System, dated February 14, 1994, or later FAA approved revisions to (1) or (2).

Imilations and Conditions: This approval should not be extended to other specific airplanes of these models on which other previously approved modifications are incorporated, unless it is determined by the installer that the interrelationship between this change and any of those other previously approved modifications will introduce no adverse effect upon the airworthiness of that airplane.

This cortificate and the supporting date which is the basis for approval shall remain in effect until surrendered, suspended, ranked, or a termination date is alterwise established by the Administrator of the

Federal Aviation Administration

Date of application: July 16, 1993

Delandremed:

Delegismano: February 16, 1994

Dolo arranded:



By direction of the Administrator

Manager, Systems & Equipment Wichita Aircraft Certification Office (Tide)

Any alteration of this certificate is punishable by a fine of not exceeding \$1,000, or imprisonment not exceeding 3 years, or both.

This certificate may be transferred in accordance with FAR 21.4

FAA Form 8110-2[10-68]

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U.S. DEPARTMENT OF TRANSPORTATION FEDERAL AVIATION ADMINISTRATION

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# NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8 DESCRIPTION OF WORK ACCOMPLISHED (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

Removed IND650 ADF indicator and installed Argus 5000 hooked to ADF 650, Garmin GPS 155, and KCS 55A. Installed ADF/Remote switch, SEL switch, and Installed as required per mfg installation manual. Installed per mfg installation manual PN 5003 Rev 3.06 dated July 8, 1994 and AC43.13-1A Chapter 2 par 21, 22 & 23.

Removed standby DG and installed Strike Finder 2000 Weather Avoidance System using KCS 55A stepper output for heading source as recommended by mfg. Installation done per mfg installation manual Drawing No 2000-10 dated Nov. 1991 and AC43.13-2A chapter 2 par 21, 22 & 23. Strike Finder ant installed per mfg installation manual and AC43.13-1A Chapter 15 Sec 6 par 842 & 846 and AC43.13-2A Chapter 3 as applicable. Weight & balance amended, logbook entry made & operator's manuals supplied.

DA L. DERITTOR ADDITIONAL SHEETS ARE ATTACHED

AU.S. GOVERNMENT PRINTING OF FICE: 1985-659-121,

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#### NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable sirvo thiness requirements

E. DESCRIPTION OF WORK ACCOMPLISHED (If more space is required, attach additional sheets. dentify with aircraft nationality and registration mark and date work completed.)

The following components were replaced:

- R.H. forward keel P/N:002-410032-30
- 2)

- Former, fuselage sta. 6.00, P/N 96-410032-58
  Former, fuselage sta. 3.00, P/N 96-410032-54
  Channel, R/H P/N 96-410032-90
  Fuselage skin, BTM Fwd R/H P/N 002-410060-19
- Nose gear door, L/H P/N 96-410024-601 Nose gear door, R/H P/N 96-410024-602 Nose cone assy. P/N 58-410015-5

Repaired Web P/N 002-410065-3 at fuselage station - 10.00. All work performed in accordance with Beechcraft model 58TC maintenance manual and AC 43.13-12.

ADDITIONAL SHEETS ARE ATTACHED

FAA AC 72-4906

±U.S. GOVERNMENT PRINTING OF FICE: 1986-659-121/40260

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FAA AIRCRAFT REGISTRY 6 - 27 - 88 CAMERA NO. I NDATE:

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. DESCRIPTION OF WORK ACCOMPLISHED (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.) Beechcraft Baron 58TC TK-25

N2130L

Installed Eleveland Conversion kit, part number 199-73, Revision B, in accordance with installation drawing 50-42. Revision C, per STC

ADDITIONAL SHEETS ARE ATTACHED

.. U S GPO 1981, 775-332 47

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D.	I certify attachm and tha	y that the repair a ents hereto have b t the information	ind/or alteration ma een made in accord furnished herein is	ade to the unit(s) identified in ite ance with the requirements of Part true and correct to the best of my	m 4 above and described o)43 of the U.S. Federal Avia knowledge.	the reverse or
ATE		7/31/87	7.	SIGNATURE OF AUTHORIZE	D INDIVIDUAL	_
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3Y	PA	A FLT, STANDARDS	MANUFACTURER	INSPECTION AUTHORIZATION	OTHER (Specify)	
	FÅ	A DESIGNEE X	REPAIR STATION	CANADIAN DEPARTMENT OF TRANSPORT INSPECTOR OF AIRCRAFT	h_{i}	
	OF AP	7/31/87	CERTIFICATE OR DESIGNATION NO 402-30	SIGNATURE OF AUTHORIZE	INDIVIDUAL	

PAA AIRCRAFT SEGIERS
CAMERA NO. 1/LOATE: 6.27.88

NOTICE

Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record. An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. DESCRIPTION OF WORK ACCOMPLISHED (If more space is required, attach additional sheets. Identify with aircraft nationality and registration mark and date work completed.)

Installed II Forrow Inc. Hodel 604 Loran Receiver in Hedio Stack at F. S. 58 and Antenna/presame on top of fuselage at F.S. 231.8 Loran system has been fround and flight checked in accordance with manufacturers Installation manual.

Loran C is placated "Loran C not approved for IFR"

All work accomplished in accordance with:

- 1) AC43.13-24 Chapter 2 Faragraph 21, 22, 23 and Chapter 3 Faragraph 36.
- 2) Nanufactures of series installation manual
- 3) AC 20-121, Appendix 1.

ADDITIONAL SHEETS ARE ATTACHED

C. FAA AC 72-4906

U S. GPO 1981 - 775-332.47

Form Approved Budget Bureau No. 04-R0058 t write in shaded areas; these

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NOTICE
Weight and balance or operating limitation changes shall be entered in the appropriate aircraft record.
An alteration must be compatible with all previous alterations to assure continued conformity with the applicable airworthiness requirements.

8. DESCRIPTION OF WORK ACCOMPLISHED (If more space is required, attach additional sheets. Identi craft nationality and registration mark and date work completed.)	ty. with air
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